

NOLTR 64-40

10335

THE FLOW FIELD BEHIND A SPHERICAL  
DETONATION IN TNT USING THE LANDAU-  
STANYUKOVICH EQUATION OF STATE FOR  
DETONATION PRODUCTS

1 COPY -2 OF 3 - 1/2  
M.C. \$ .200  
R.O.F. M.E. \$ .650

43 P

NOL  
10 DECEMBER 1964  
UNITED STATES NAVAL ORDNANCE LABORATORY, WHITE OAK, MARYLAND

NOLTR 64-40

DDC  
PUBLISHED  
FFB 1 1965  
DDC IRA C

ARCHIVE COPY

THE FLOW FIELD BEHIND A SPHERICAL DETONATION IN TNT USING THE LANDAU-STANYUKOVICH EQUATION OF STATE FOR DETONATION PRODUCTS

Prepared by:  
M. Lutzky

ABSTRACT: Calculations have been made of the flow field in the isentropic region behind a detonation wave in TNT, using the Landau-Stanyukovich equation of state for the detonation products (as described by Zeldovich and Kompaneets). Adjustable constants in this equation have been evaluated by imposing ideal gas behavior on the detonation products in the large expansion (low density) limit, and by fitting to an experimental curve of detonation velocity versus loading density. Calculated values of Chapman-Jouguet variables correspond fairly well with experimental values at various loading densities, with the exception of the temperatures, which seem to be far too low. This is connected with the fact that the theory predicts an upper limit to the loading density at which an explosive will detonate; at this point the thermal energy vanishes and only the elastic energy contributes to the energy of detonation.

PUBLISHED FEBRUARY 1965

Air-Ground Explosions Division  
Explosions Research Department  
U. S. NAVAL ORDNANCE LABORATORY  
WHITE OAK, MARYLAND

NOLTR 64-40

10 December 1964

THE FLOW FIELD BEHIND A SPHERICAL DETONATION IN TNT USING THE LANDAU-STANYUKOVICH EQUATION OF STATE FOR DETONATION PRODUCTS

Calculations of the airshock motion produced by a spherical TNT explosion, with the reaction products considered to be gaseous, have given satisfactory agreement with experimental results. However, the experimental motion of the explosive interface and the second shock have not agreed with the theoretical calculations. An attempt to clarify these discrepancies has led to consideration of the Landau-Stanyukovich solid state model for the reaction products of a condensed explosive. The Landau-Stanyukovich equation of state has been utilized to calculate the flow field in the reaction products behind the Chapman-Jouguet zone - the so-called Taylor Wave distribution - and the results are presented in this report. Preliminary determinations of this distribution have already been used as initial conditions for the calculation of the subsequent explosion motion, and have been reported elsewhere.

Support for this investigation has been provided by the Defense Atomic Support Agency under Nuclear Weapons Effects Research Subtask 01.002 (NOL-428).

This report has been approved for open publication by the Department of Defense, Office of Assistant Secretary of Defense (Public Affairs).

R. E. ODENING  
Captain, USN  
Commander

*I. Kabik*

I. KABIK  
By direction

## CONTENTS

	Page
INTRODUCTION.....	1
THE LSZK EQUATION OF STATE.....	2
ISENTROPIC PROCESSES.....	4
CHAPMAN-JOUQUET CONDITIONS.....	6
EVALUATION OF PARAMETERS.....	9
FLOW FIELD BEHIND DETONATION SHOCK.....	11
CONCLUDING REMARKS.....	13
ACKNOWLEDGMENT.....	14
REFERENCES.....	15

## TABLES

Table	Title
1	Comparison of Detonation Velocities Calculated for LSZK Substance with Detonation Velocities Determined at Bruceton
2	Detonation Parameters Calculated with LSZK Equation of State, for TNT, $Q = 1018 \text{ cal/gm}$
3	Experimental Values for Detonation Parameters of TNT (Dremin, et al)
4	Detonation Wave for TNT ( $\rho_0 = 1.625 \text{ gm/cc}$ )
5	Detonation Wave for TNT ( $\rho_0 = 1.59 \text{ gm/cc}$ )
6	Detonation Wave for TNT ( $\rho_0 = 1.45 \text{ gm/cc}$ )
7	Detonation Wave for TNT ( $\rho_0 = 1.30 \text{ gm/cc}$ )
8	Detonation Wave for TNT ( $\rho_0 = 1.14 \text{ gm/cc}$ )
9	Detonation Wave for TNT ( $\rho_0 = 1.00 \text{ gm/cc}$ )

## INTRODUCTION

The theory of the detonation process for a  $\gamma$ -law gas, whose detonation product is also a  $\gamma$ -law gas, has been quite completely worked out and is available in many places.<sup>1,2</sup> In particular, the conditions at the Chapman-Jouguet state can be derived, and it can be shown that the detonation velocity is a function only of the heat of detonation and the  $\gamma$  for the detonation products, for sufficiently large detonation pressures. In addition, differential equations have been derived for the flow behind the detonation wave, and have been solved for certain explosives and geometries.<sup>2,3,4</sup>

The theory of the detonation of a solid explosive, on the other hand, is in a much less satisfactory state. Experiments<sup>5,6</sup> have shown that the detonation velocity of a solid explosive depends on the initial density, unlike the detonation velocity of gaseous detonations. Furthermore, the explosion products of condensed explosives are obtained at pressures of the order of megabars, and at densities approaching 2 grams/cm<sup>3</sup>, under which conditions their behavior becomes extremely complex. Consequently, various attempts have been made to find an equation of state for the explosion products by treating the highly compressed gas as a solid.

The first such attempt is due to H. Jones,<sup>7</sup> who developed an equation of the Grüneisen type, based on the Einstein model of a solid, of the form  $p = Ae^{-av} - B + fRT$ , where a, A, B and f are constants. The equation of state which we consider in this paper, however, was derived by Landau and Stanyukovich,<sup>8,9</sup> who also approached the problem by drawing an analogy between the state of the detonation products of a condensed explosive and the crystal lattice of the solid state. It is well known that the energy of a solid body has a two-fold origin: it is made up of an elastic energy arising from the binding forces between the atoms and molecules and a thermal energy connected with oscillations of the atoms or molecules about their positions of stable equilibrium. Landau and Stanyukovich have attempted to describe the behavior of the detonation product by considering it as a solid with the property that the elastic energy and the elastic part of the pressure are predominant. The theory has been described and expanded by Zeldovitch and Kompaneets,<sup>8</sup> so that we refer to it as the LSZK theory. The purpose of this paper is to make some computations using the LSZK equations of state, and, in particular, to calculate the flow field behind the detonation shock in a condensed explosive.

## THE LSZK EQUATION OF STATE

For the sake of completeness, we present here a description of the LSZK equation of state, and a derivation of some of its properties.

The LSZK equation of state may be written<sup>8</sup>

$$P = \frac{B}{v^\gamma} + \frac{C_{V1} \left( \frac{\gamma}{2} - \frac{1}{6} \right)}{v} T \quad (1)$$

$$E = \frac{B}{(\gamma-1)v^{\gamma-1}} + C_V T, \quad (2)$$

where  $P$  = pressure

$E$  = energy density (per unit mass)

$v$  = specific volume

$T$  = temperature

and  $B$ ,  $C_{V1}$ ,  $C_V$  and  $\gamma$  are constants.  $\gamma$  is a dimensionless constant serving as a polytropic index connected with the intermolecular forces;  $C_V$  is the specific heat at constant volume;  $C_{V1}$  is a specific heat associated with the appropriate lattice vibrations; and  $B$  is a constant having the units  $(\frac{\text{gm}}{\text{cm}^3})^{1-\gamma} \frac{\text{calories}}{\text{gram}}$ . The elastic part of the pressure is  $\frac{B}{v^\gamma}$ , and  $\frac{B}{(\gamma-1)v^{\gamma-1}}$  is the elastic part of the energy.

Eliminating  $T$  between (1) and (2), we obtain the expression

$$P = \frac{E}{\alpha v} + \frac{B}{v^\gamma} \left\{ 1 - \frac{1}{\alpha(\gamma-1)} \right\}, \quad (3)$$

where

$$\alpha = \frac{C_V}{C_{V1}} \left\{ \frac{1}{\frac{\gamma}{2} - \frac{1}{6}} \right\}. \quad (4)$$

$\alpha$  is a convenient variable which will be used in this report. In terms of  $\alpha$ , (1) and (2) may be written:

$$P = \frac{B}{v^\gamma} + \frac{C_V T}{\alpha v} \quad (5)$$

$$E = \frac{B}{(\gamma-1)v^{\gamma-1}} + C_v T \quad (6)$$

Another convenient parameter which we will find useful is the quantity  $y$ , defined as the ratio of the thermal part of the pressure to the elastic part:

$$y = \frac{(C_v T / \alpha v)}{(B/v)} = \frac{C_v T}{\alpha B} v^{\gamma-1} \quad (7)$$

Clearly, (5) and (6) may now be written in the form:

$$P = \frac{B}{v^\gamma} (1 + y) \quad (8)$$

$$E = \frac{B\alpha}{v^{\gamma-1}} \left\{ y + \frac{1}{\alpha(\gamma-1)} \right\} \quad (9)$$

## ISENTROPIC PROCESSES

It is possible to obtain an expression for the pressure of the form  $P = P(\rho)$ , valid for isentropic processes of an LSZK substance, by combining equation (3) with

$$dE = \frac{P}{\rho} d\rho , \quad (10)$$

which is the differential equation of an isentropic process. Differentiating (3), we obtain

$$dP = \frac{1}{\alpha} (\rho dE + E d\rho) + B \gamma \rho^{\gamma-1} \left\{ 1 - \frac{1}{\alpha(\gamma-1)} \right\} d\rho \quad (11)$$

Using (10) to eliminate  $dE$  from (11), we obtain

$$dP = \frac{1}{\alpha} \left( \frac{P}{\rho} d\rho + E d\rho \right) + B \gamma \rho^{\gamma-1} \left\{ 1 - \frac{1}{\alpha(\gamma-1)} \right\} d\rho \quad (12)$$

Solving (3) for  $E$  and substituting into (12), we obtain the differential equation

$$\frac{dP}{d\rho} - \left( \frac{1+\alpha}{\alpha} \right) \frac{P}{\rho} = B \rho^{\gamma-1} \left( \gamma - 1 - \frac{1}{\alpha} \right) \quad (13)$$

which has the solution:

$$P(\rho) = K \rho^{\frac{1+\alpha}{\alpha}} + B \rho^{\gamma} \quad (14)$$

where  $K$  is a constant of integration. We are now in a position to obtain expressions for  $E$ , the sound speed  $c$ , the temperature  $T$ , etc. as functions of density alone, valid for isentropic processes of an LSZK substance. Thus, putting (14) into (3), and solving for  $E$ , we obtain:

$$E(\rho) = \alpha K c \frac{1}{\alpha} + \left( \frac{1}{\gamma-1} \right) B \rho^{\gamma-1} \quad (15)$$

Similarly,

$$T = \frac{\gamma K_p}{C_v}^{\frac{1}{\gamma}} \quad (16)$$

and

$$c^2 = K \left( \frac{1 + \alpha}{\alpha} \right) \rho^{\frac{1}{\alpha}} + B_v \rho^{\gamma-1} \quad (17)$$

## CHAPMAN-JOUQUET CONDITIONS

We now obtain the initial conditions at an LSZK detonation, in terms of the variable  $y$ . We consider that the detonation wave consists of a shock traveling at speed  $D$ , followed immediately by a region of isentropic expansion. The region of chemical reaction behind the shock is considered to be infinitely thin. Values of the hydrodynamic parameters in the undetonated explosive ahead of the shock are given a subscript  $0$  so that  $v_0$  is the specific volume of the original explosive. We first obtain  $\frac{v}{v_0}$  as a function of  $y$ . From the Rankine-Hugoniot (R-H) relations at the shock, we have

$$\frac{v_0}{v} = \frac{D}{D-u} ; \quad (18)$$

where  $u$  = particle velocity  
 $D$  = detonation velocity.

Using the detonation property  $D = u + c$  (19)

equation (18) becomes

$$\frac{v_0}{v} = \frac{u}{c} + 1. \quad (20)$$

Another R-H relation yields

$$P = Du/v_0 \quad (21)$$

which may be written in the form

$$\left(\frac{u}{c} + 1\right) \frac{u}{c} = \frac{Pv_0}{c^2} . \quad (22)$$

Using  $P = B_0^Y (1 + y)$  and the equation (20), (22) may be put into the form

$$\frac{v_0}{v} - 1 = \frac{B_0^Y (1 + y)}{c^2} . \quad (23)$$

$c^2$  may be obtained as a function of  $\rho$  and  $y$  by eliminating  $K$  between (14) and (17), and then using  $P = B\rho^Y (1 + y)$  to eliminate  $P$ ; the result is

$$c^2 = B\rho^{Y-1} \left( Y + \left\{ \frac{1 + \alpha}{\alpha} \right\} y \right) . \quad (24)$$

Inserting (24) into (23) we obtain  $\frac{v_0}{v}$  as a function of  $y$ :

$$\frac{v_0}{v} = 1 + \frac{(1 + y)}{Y + \left\{ \frac{1 + \alpha}{\alpha} \right\} y} . \quad (25)$$

The next step is to obtain  $v_0$  itself as a function of  $y$ ; this will enable us to solve for the parameter  $y$  as a function of the known quantity  $v_0$ .

We write the R-H equation for the energy in the form

$$E = Q + \frac{1}{2} P (v_0 - v) = Q + \frac{1}{2} Pv \left( \frac{v_0}{v} - 1 \right) , \quad (26)$$

where  $Q$  is the chemical energy released by each gram of explosive; and using (25), this becomes:

$$E = Q + \frac{1}{2} Pv \left( \frac{1 + y}{Y + \left\{ \frac{1 + \alpha}{\alpha} \right\} y} \right) . \quad (27)$$

Eliminating  $P$  by using  $P = B\rho^Y (1 + y)$  we obtain

$$\frac{E}{Q} = 1 + \frac{B}{Q} \frac{(1 + y)^2}{2v^{Y-1}} \left( \frac{1}{Y + y \left\{ \frac{1 + \alpha}{\alpha} \right\}} \right) . \quad (28)$$

Another expression for  $E/Q$  may be obtained from equation (9):

$$\frac{E}{Q} = \frac{B}{Q} - \frac{\alpha}{v^{Y-1}} \left\{ y + \frac{1}{\alpha(Y-1)} \right\} . \quad (29)$$

Equating (28) and (29), and solving for  $v$ , we obtain

$$v^{\gamma-1} = \frac{B}{Q} \left( \alpha y + \frac{1}{\gamma-1} - \frac{(1+y)^2}{2(\gamma + \left\{ \frac{1+\alpha}{\gamma} \right\} y)} \right) . \quad (30)$$

Eliminating  $v$  between (30) and (25), we obtain the expression:

$$v_0 = \left( \frac{B}{Q} \right)^{\gamma-1} \left[ \alpha y + \frac{1}{\gamma-1} - \frac{(y+1)^2}{2w} \right]^{\frac{1}{\gamma-1}} \left( 1 + \frac{1+y}{w} \right) , \quad (31)$$

where

$$w = y + \left( \frac{1+\alpha}{\alpha} \right) y . \quad (32)$$

Since  $v_0$ , the specific volume of the solid explosive, is a known quantity, we may solve (31) for  $y$ , by an iterative process. Since  $v$  is a known function of  $y$ , by virtue of (30), we can find  $P$  by using the expression  $P = \frac{B}{v^{\gamma}} (1+y)$ ;  $E$  may be found from (28) or (29); and  $c^2$  may be found from equation (24).

The particle velocity at the front,  $u$ , may be found from (20), and is given by

$$u = \frac{c(1+y)}{\gamma + \left\{ \frac{1+\alpha}{\alpha} \right\} y} . \quad (33)$$

Finally, the detonation velocity can be found from

$$D = u + c = c \left[ 1 + \frac{1+y}{\gamma + \left\{ \frac{1+\alpha}{\alpha} \right\} y} \right] . \quad (34)$$

Thus, the detonation velocity is seen to be a function of  $v_0$ , the initial specific volume, corresponding to the well-known experimental result for solid explosives.

## EVALUATION OF PARAMETERS

The three undetermined parameters,  $\gamma$ ,  $\alpha$ , and  $B/Q$ , which appear in the LSZK equation of state, must be evaluated by using experimental data. It can be seen from equation (14), which describes the isentropic  $P-\rho$  relation for an LSZK substance, that if  $\frac{1+\alpha}{\alpha} < \gamma$ ,  $P(\rho)$  approaches

$K_p \frac{1+\alpha}{\alpha}$  as  $\rho$  approaches zero. We assume that in the limit of low pressures the detonation products behave as ideal gases, with a constant value of the specific heat ratio, denoted here by  $\kappa$ . (A reasonable value for  $\kappa$  seems to be 1.34, obtainable by averaging the gammas for the various gaseous constituents according to the composition of the products at low pressure.) It is thus clear that in order to obtain the correct behavior of the detonation products at low pressures, we must set  $\alpha = \frac{1}{\kappa-1}$ .

The remaining constants may be evaluated by referring to the experimental results for the dependence of the detonation velocity on the density. After a particular value is assigned for  $\gamma$  ( $\gamma_{\text{ex}}$ ), a series of values for  $B/Q$  may be obtained by carrying out a point by point comparison of the theoretical plot of  $\ln D$  vs.  $(\ln \rho_0 + \frac{1}{\gamma-1} \ln \frac{B}{Q})$  (obtainable from equations (31) and (34)) with the experimental plot of  $\ln D$  vs.  $\ln \rho_0$ . Since  $B/Q$  must be a constant, the accuracy of the fit is determined by the amount of variation in the values of  $B/Q$  obtained, and  $\gamma$  may be adjusted to make this variation a minimum.

This process has been carried out for TNT, using an empirical relation between detonation velocity and loading density determined at the Explosives Research Laboratory, at Bruceton.<sup>8</sup> This relation may be written  $D = 0.1785 + 0.3225 \rho_0$ , where  $D$  is in centimeters per microsecond and  $\rho_0$  is in grams per cubic centimeter. Using  $\kappa = 1.34$ , and with the heat of detonation chosen<sup>11</sup> to be 1018 cal/gm, the results are  $\gamma = 2.78$ ,  $B/Q = 0.53562$  and  $\alpha = 2.9412$ .

Equations (31) and (34) may now be utilized to provide the dependence of the detonation velocity on the loading density, by letting the parameter  $\gamma$  run through a range of values. Table 1 gives a comparison of this theoretical curve with the empirical relationship, and it can be seen that the fit is quite good.

It is interesting to note that this formalism predicts an upper density limit to the detonability, at the loading density  $\rho_0 = 1.793$  gm/cc and detonation velocity  $D = 0.757$  cm/usec. This comes about because of the fact that at this point the value of the parameter  $\gamma$  is zero ( $\gamma$  decreases with increasing loading density), and  $\gamma$  cannot be negative, because of its physical meaning as a ratio of pressures (see equation (7)).

In fact, equation (7) implies that at this limiting point, the only contribution to the pressure comes from the elastic part, while the thermal pressure vanishes. Zeldovich and Kompaneets<sup>8</sup> have the following to say about the physical significance of this phenomenon: "It is possible to have charge densities for which the thermal energy is much smaller than the elastic part. This corresponds to  $\gamma$  being nearly zero....It is not quite clear what happens when the charge density is large. It can be assumed that in this case the dissociation reaction does not go to completion, since the supply of chemical energy is insufficient for overcoming the work required by the elastic repulsion forces between the molecules. It appears as though the chemical energy does not suffice for the molecular rearrangement which leads to an explosion." This prediction is especially interesting inasmuch as it is known that TNT exhibits increasing resistance to detonation with increasing loading density, as, in fact, do most solid explosives.

Now that values for the constants in the LSZK equation have been arrived at, the conditions at the Chapman-Jouguet state may be computed by using the expressions developed in the preceding section. This has been done for TNT at several loading densities and the results are presented in Table 2. (The temperatures were calculated using  $C_v = 0.3$  cal/gm-degree.) Table 3 presents experimental values for the C-J state, determined by Dremin, et al;<sup>12</sup> the correspondence between calculated values and experiment appears good, except for the temperatures, which seem to be far below the generally accepted values for detonation temperatures of several thousand degrees.

The fact that the ratio of elastic pressure to total pressure increases as the loading density increases may be verified in the last column of Table 2. For instance, for  $\rho_0 = 1.625$ , this ratio is 0.974, which means that, for the isentrope given in equation (14), 97.4% of the pressure comes from the elastic pressure term. For  $\rho_0 = 1.00$  gm/cc, this ratio is 0.805.

Consequently, in the vicinity of the Chapman-Jouguet state, the LSZK isentrope may be approximated by a polytropic relation, with exponent equal to 2.78. In this connection, it is interesting to note the experimental results of Deal,<sup>6</sup> which indicate that the explosion products isentrope for RDX-TNT may be fitted quite closely to a polytropic  $P-\rho$  relation, with  $\gamma = 2.77$ , at least down to 500 bars. It thus seems likely that the LSZK equation of state for TNT not only yields the proper  $D$  vs  $\rho_0$  relationship, but also provides the proper isentrope, both in the vicinity of the Chapman-Jouguet state and in the large expansion limit of low pressure and density.

## FLOW FIELD BEHIND DETONATION SHOCK

The isentropic flow behind a detonation shock in a spherical explosive is governed by the differential equations:

$$\frac{du}{d\xi} = \frac{2uc^2}{\xi \{ (u - \xi)^2 - c^2 \}} \quad (36)$$

$$\frac{dc^2}{d\xi} = \frac{2uc^2 (\xi - u)}{\{ (u - \xi)^2 - c^2 \} \xi} f, \quad (37)$$

where  $u$  = particle velocity

$r$  = radial distance of detonation shock from origin

$\xi = \frac{r}{t}$ ,  $t$  = time

$c$  = sound speed

$$f = \left( \frac{\rho}{dP/d\rho} \right) \frac{d^2 P}{d\rho^2}, \text{ where } \rho = \text{density} = \frac{1}{v}. \quad (38)$$

Calculating  $f$  by means of (14), we obtain:

$$f = \frac{\frac{K(1+\alpha)}{\alpha} + BY(\gamma-1)\rho^{\gamma-1-\frac{1}{\alpha}}}{\frac{K(1+\alpha)}{\alpha} + BY\rho^{\gamma-1-\frac{1}{\alpha}}}. \quad (39)$$

To utilize (39) in the system of differential equations (36), (37), it is necessary to express  $f$  as a function of  $c^2$ . This may be done (in principle) by solving (17) for  $\rho$  in terms of  $c^2$ , and substituting the result into (39). Unfortunately, it is not possible to invert (17) analytically, in closed form, so that an alternative approach must be used. The procedure chosen here is to convert equations (36) and (37) into a set in which the dependent variables are  $u$  and  $\rho$ , rather than  $u$  and  $c^2$ . In this case, we can use  $f$  in the form (39), as a function of  $\rho$ . To effect this change of variable we make use of the equation

$$\frac{dc^2}{d\xi} = \frac{dc^2}{d\rho} \frac{d\rho}{d\xi} \quad (40)$$

Since  $c^2 = dP/d\rho$  in the isentropic flow, we may put  $dc^2/d\rho = d^2P/d\rho^2$  and using (17) we get:

$$\frac{dc^2}{d\rho} = \frac{K}{\alpha} \left( \frac{1+\alpha}{\alpha} \right) \rho^{\frac{1-\alpha}{\alpha}} + BY (\gamma - 1) \rho^{\gamma-2} \quad (41)$$

Consequently,  $\frac{dc^2}{d\xi} = \left\{ \frac{K}{\alpha} \left( \frac{1+\alpha}{\alpha} \right) \rho^{\frac{1-\alpha}{\alpha}} + BY (\gamma - 1) \rho^{\gamma-2} \right\} \frac{d\rho}{d\xi}$ , and the differential equations become:

$$\frac{du}{d\xi} = \frac{2uc^2}{\xi \{ (u - \xi)^2 - c^2 \}} \quad (42)$$

$$\frac{d\rho}{d\xi} = \frac{2uc^2 (\xi - u)}{\{ (u - \xi)^2 - c^2 \} \xi} \left\{ \frac{f(\rho)}{\frac{K}{\alpha} \left( \frac{1+\alpha}{\alpha} \right) \rho^{\frac{1-\alpha}{\alpha}} + BY (\gamma - 1) \rho^{\gamma-2}} \right\}, \quad (43)$$

where  $c^2 = K \left( \frac{1+\alpha}{\alpha} \right) \rho^{\frac{1}{\alpha}} + BY \rho^{\gamma-1}$ , and  $f(\rho)$  is given by (39). These equations are to be solved subject to the conditions  $u = u_D$ ,  $\rho = \rho_D$  at  $\xi = D$ , where

$D$  = detonation velocity

$u_D$  = particle velocity at the detonation shock

$\rho_D$  = density at the detonation shock.

$u_D$  and  $\rho_D$  may be found from (33) and (25), after  $\gamma$  has been found from equation (31).

These calculations have been carried out for TNT on an IBM-7090 electronic computer, for the loading densities 1.625, 1.59, 1.45, 1.30, 1.14, and 1.00 gm/cc. The results are given in Tables 4-9. The first column is a dimensionless distance, the radius of the original charge being taken as the unit. Pressures are in megabars, velocities in centimeters per microsecond, energy densities in megabar-cc per gram and densities in grams per cubic centimeter. It will be seen that the parameters vary in the well-known way first demonstrated by Taylor,<sup>9</sup> with the region of constant state surrounding the origin.

## CONCLUDING REMARKS

Up until now, very little has been said about the temperature. To evaluate this quantity, we must know the value of  $C_V$ , which is not determined by the other constants. (Only the ratio  $C_V/C_{V1}$  is determined by equation (4).) If  $C_V$  is taken to be 0.3 cal/gm, an approximate average value for detonation products, the C-J temperature (for  $\rho_0 = 1.625$  gm/cc) turns out to be 582.9°K, which seems to be too low. This is connected with the phenomenon of the decreasing importance of the thermal pressure with increasing loading density, which was mentioned above. Though this phenomenon is consistent with the known resistance to detonation of TNT at high densities, and with the experimental results of Deal,<sup>9</sup> it is not yet certain whether it is a real effect or whether it is a result of the incompleteness of the LSZK theory\*. In any case, it is believed that in all applications where the temperature is not needed, and only an  $(E, p, v)$  equation of state is required (such as the calculation of the non-reactive, isentropic expansion of detonation products by means of hydrodynamic computer codes), the LSZK equation of state (in particular, equation (3)) may be used with confidence.

It is probably not possible to decide on the correctness of the LSZK equation of state by experimental observations of the detonation process alone. A possible approach is to use the results for the distribution behind the detonation as initial conditions for a hydrodynamic code computation of the detonation of a sphere of TNT in air, using the LSZK equation of state for the expanding detonation products. The motion of the second shock through the product gases is expected to be a sensitive function of the equation of state used, and one can attempt to compare the calculated results with the evidence obtained from photographic records. The behavior of the air shock, though a much less sensitive function of the equation of state for explosion products, might also provide a useful check.

Preliminary hydrodynamic calculations have already been carried out on an IBM-7090 and are reported elsewhere;<sup>10</sup> more refined computations are in progress at the present time.

---

\* Jacobs<sup>13</sup> has pointed out that a reinterpretation of the partition between elastic and thermal energy leads to a theory which does not involve a limiting density, or vanishing thermal pressure. This theory retains the LSZK form for the equation of state, but does not make use of Zeldovich's<sup>9</sup> arguments for the physical meaning of the constants  $C_{V1}$ ,  $C_{V2}$ , and  $C_V$ .

ACKNOWLEDGMENT

The author gratefully acknowledges fruitful and enlightening discussions with L. Rudlin, S. J. Jacobs, H. M. Sternberg, H. Hurwitz, and J. W. Enig.

## REFERENCES

1. Penner, S. S. and Mullins, B. P., Explosions, Detonations, Flammability and Ignition, Pergamon Press, 1959, Chapter 5
2. Landau, L. D. and Lifshitz, E. M., Fluid Mechanics, Pergamon Press, 1959 (Addison-Wesley Publishing Co., Inc.) Chapter 14
3. Taylor, G. I., The Dynamics of the Combustion Products Behind Plane and Spherical Detonation Fronts in Explosives, Proc. Roy. Soc. A200, 1061, Pgs 235-247 (1950)
4. Lutzky, M., The Spherical Taylor Wave for the Gaseous Products of Solid Explosives, NAVWEPS Report 6848 (1961)
5. Seal, W. E., Measurement of the Reflected Shock Hugoniot and Isentrope for Explosion Reaction Products, Physics of Fluids, 1, 6, P. 523 (1958)
6. MacDougall, D. P., Messerly, G. H., Hurwitz, M. D., et al, "The Rate of Detonation of Various Explosive Compounds and Mixtures," OSRD-5611. See also: Urizar, M. J., James Jr., E., Smith, L. C., Detonation Velocity of Pressed TNT, Physics of Fluids, 4, 2, P. 262 (1961)
7. Jones, H., 1941. See Cole, R. H., Underwater Explosions, (Princeton University Press, 1948)
8. Landau, L. D. and Stanyukovich, K. P., On the Study of Detonation in Condensed Explosives, Doklady Akad. Nauk SSSR 46, 399 (1945)
9. Zeldovich, Ia.B., and Kompaneets, A. S., Theory of Detonation, Academic Press (1960), Chapter 14
10. Rudlin, L., On the Origin of Shockwaves from Spherical Condensed Explosions in Air, U. S. Naval Ordnance Laboratory NOLTR 63-220, Part 3, Appendix B (to be published)
11. Rudlin, L., An Approximate Solution of the Flow Within the Reaction Zone Behind a Spherical Detonation Wave in TNT, U. S. Naval Ordnance Laboratory, NAVWEPS Report 7364, April 1960
12. Dremin, A. N., Zaitsev, V. M., Ilyukhin, V. S., Pokhil, P. r., Detonation Parameters, Eighth Symposium (International) on Combustion, Williams and Wilkins Co., Baltimore, P. 610 (1962)
13. Jacobs, S. J., A New Interpretation of the Zeldovich-Kompaneets Treatment of the Equation of State for Detonation Products, U. S. Naval Ordnance Laboratory Internal Memorandum, 5 June 1954

Table 1

Comparison of Detonation Velocities Calculated for  
LSZK Substance with Detonation Velocities Determined at Bruceton

$\rho_0$ ( $\frac{\text{gm}}{\text{cc}}$ )	$D(\frac{\text{cm}}{\text{usec}})$ ; LSZK	$D(\frac{\text{cm}}{\text{usec}})$ ; (Bruceton)
1.7935	0.7572	0.7569
1.6620	0.7146	0.7145
1.5535	0.6795	0.6795
1.4412	0.6433	0.6433
1.3655	0.6189	0.6189
1.2995	0.5977	0.5976
1.2412	0.5791	0.5788
1.1773	0.5588	0.5582
1.1320	0.5444	0.5436
1.1009	0.5345	0.5335
1.0034	0.5039	0.5021
0.9590	0.4900	0.4878
0.9256	0.4797	0.4770
0.9010	0.4720	0.4691
0.8565	0.4584	0.4547
0.8082	0.4437	0.4391
0.7703	0.4322	0.4269
0.7331	0.4211	0.4149

Table 2

Detonation Parameters Calculated with  
LSZK Equation of State, for TNT,  $Q = 1018 \text{ cal/gm}$

$\rho_0 \text{ (gm/cc)}$	P(Kbars)	E( $\frac{\text{megabar-cc}}{\text{gram}}$ )	$\rho \text{ (gm/cc)}$	$u \text{ (cm/usec)}$	$D \text{ (cm/usec)}$	T(degree Kelvin)	$\frac{P(\text{elastic})}{P(\text{total})}$
1.625	214.3	0.06022	2.217	0.188	0.703	582.9	0.974
1.59	203.5	0.05973	2.171	0.185	0.691	698.4	0.968
1.45	163.8	0.0579	1.988	0.175	0.646	1141.7	0.941
1.30	127.6	0.05607	1.792	0.164	0.598	1582.5	0.905
1.14	95.4	0.05431	1.583	0.153	0.547	2013.3	0.857
1.00	72.2	0.05293	1.400	0.144	0.503	2356.7	0.805

Table 3

Experimental Values for Detonation Parameters  
of TNT (Dremin, et al)

$\rho_0$ ( $\frac{\text{gm}}{\text{cc}}$ )	P(kbars)	$u$ ( $\frac{\text{cm}}{\text{usec}}$ )	$D$ ( $\frac{\text{cm}}{\text{usec}}$ )
1.59	202	0.183	0.694
		0.180	
1.45	162	0.172	0.650
		0.168	
1.30	123	0.158	0.600
		0.156	
1.14	96	0.145	0.557
		0.142	
1.00	64	0.130	0.510
		0.129	

## NOLTR 64-40

Table 4

Detonation Wave for TNT ( $\rho_0 = 1.625$  gm/cc)

DISTANCE X/RADILS	VELOCITY CM/USEC	DENSITY GM/CC	PRESSURE MEGABARS	ENERGY DENS (J/CG-CC)/GM
0.	0.	1.29163E 00	4.91537E-02	2.63062E-02
4.57265E-02	0.	1.29163E 00	4.91537E-02	2.63062E-02
9.14531E-02	0.	1.29163E 00	4.91537E-02	2.63062E-02
1.37180E-01	0.	1.29163E 00	4.91537E-02	2.63062E-02
1.82906E-01	0.	1.29163E 00	4.91537E-02	2.63062E-02
2.28633E-01	0.	1.29163E 00	4.91537E-02	2.63062E-02
2.74359E-01	0.	1.29163E 00	4.91537E-02	2.63062E-02
3.20086E-01	0.	1.29163E 00	4.91537E-02	2.63062E-02
3.65812E-01	0.	1.29163E 00	4.91537E-02	2.63062E-02
4.11539E-01	0.	1.29163E 00	4.91537E-02	2.63062E-02
4.57265E-01	5.62295E-05	1.29163E 00	4.91537E-02	2.63062E-02
4.85730E-01	2.86409E-03	1.30326E 00	5.03598E-02	2.66499E-02
5.14195E-01	6.72565E-03	1.31989E 00	5.21153E-02	2.71453E-02
5.42659E-01	1.11811E-02	1.33975E 00	5.42632E-02	2.77425E-02
5.71124E-01	1.60719E-02	1.36220E 00	5.67594E-02	2.84252E-02
5.99589E-01	2.13167E-02	1.38690E 00	5.95935E-02	2.91858E-02
6.28053E-01	2.68724E-02	1.41367E 00	6.27591E-02	3.00208E-02
6.56518E-01	3.27201E-02	1.44242E 00	6.62796E-02	3.09299E-02
6.84983E-01	3.88586E-02	1.47311E 00	7.01775E-02	3.19152E-02
7.13447E-01	4.53024E-02	1.50582E 00	7.44899E-02	3.29814E-02
7.41912E-01	5.20822E-02	1.54067E 00	7.92682E-02	3.41357E-02
7.70377E-01	5.92466E-02	1.57788E 00	8.45828E-02	3.53900E-02
7.98841E-01	6.68685E-02	1.61777E 00	9.05303E-02	3.67566E-02
8.27306E-01	7.50551E-02	1.66085E 00	9.72468E-02	3.82610E-02
8.55771E-01	8.39676E-02	1.70785E 00	1.04932E-01	3.99349E-02
8.84235E-01	9.38604E-02	1.75995E 00	1.13896E-01	4.18301E-02
9.00523E-01	1.00118E-01	1.79278E 00	1.19789E-01	4.30454E-02
9.15477E-01	1.06375E-01	1.82544E 00	1.25843E-01	4.42709E-02
9.29105E-01	1.12632E-01	1.85790E 00	1.32050E-01	4.55045E-02
9.41417E-01	1.18890E-01	1.89010E 00	1.38399E-01	4.67442E-02
9.52435E-01	1.25147E-01	1.92200E 00	1.44880E-01	4.79877E-02
9.62185E-01	1.31405E-01	1.95357E 00	1.51482E-01	4.92332E-02
9.70701E-01	1.37662E-01	1.98476E 00	1.58192E-01	5.04786E-02
9.78022E-01	1.43919E-01	2.01555E 00	1.65000E-01	5.17220E-02
9.84193E-01	1.50177E-01	2.04590E 00	1.71892E-01	5.29615E-02
9.89262E-01	1.56434E-01	2.07578E 00	1.78856E-01	5.41953E-02
9.93282E-01	1.62691E-01	2.10517E 00	1.85880E-01	5.54217E-02
9.96309E-01	1.68949E-01	2.13405E 00	1.92952E-01	5.66392E-02
9.98398E-01	1.75206E-01	2.16240E 00	2.00060E-01	5.78461E-02
9.99609E-01	1.81463E-01	2.19020E 00	2.07191E-01	5.90411E-02
1.00000E 00	1.87721E-01	2.21743E 00	2.14333E-01	6.02228E-02

## WOLTR 64-40

Table 5

Detonation Wave for TNT ( $\rho_0 = 1.59 \text{ gm/cc}$ )

DISTANCE X/RADILS	VELOCITY CM/USEC	DENSITY GM/CC	PRESSURE MECABARS	ENERGY DENS (MEG-CC)/GM
0.	0.	1.26271E 00	4.67753E-02	2.67108E-02
4.56064E-02	0.	1.26271E 00	4.67753E-02	2.67108E-02
9.12129E-02	0.	1.26271E 00	4.67753E-02	2.67108E-02
1.36819E-01	0.	1.26271E 00	4.67753E-02	2.67108E-02
1.82426E-01	0.	1.26271E 00	4.67753E-02	2.67108E-02
2.28032E-01	0.	1.26271E 00	4.67753E-02	2.67108E-02
2.73639E-01	0.	1.26271E 00	4.67753E-02	2.67108E-02
3.19245E-01	0.	1.26271E 00	4.67753E-02	2.67108E-02
3.64851E-01	0.	1.26271E 00	4.67753E-02	2.67108E-02
4.10458E-01	0.	1.26271E 00	4.67753E-02	2.67108E-02
4.56064E-01	9.91348E-05	1.26271E 00	4.67753E-02	2.67108E-02
4.84596E-01	2.72887E-03	1.27354E 00	4.78596E-02	2.70294E-02
5.13127E-01	6.52934E-03	1.28980E 00	4.95179E-02	2.75111E-02
5.41659E-01	1.09217E-02	1.30925E 00	5.15509E-02	2.80933E-02
5.70191E-01	1.57492E-02	1.33128E 00	5.39162E-02	2.87595E-02
5.98722E-01	2.09282E-02	1.35553E 00	5.66005E-02	2.95019E-02
6.27254E-01	2.64154E-02	1.38182E 00	5.96061E-02	3.03171E-02
6.55785E-01	3.21914E-02	1.41005E 00	6.29463E-02	3.12047E-02
6.84317E-01	3.82546E-02	1.44021E 00	6.66453E-02	3.21667E-02
7.12848E-01	4.46192E-02	1.47235E 00	7.07384E-02	3.32074E-02
7.41380E-01	5.13151E-02	1.50660E 00	7.52742E-02	3.43340E-02
7.69912E-01	5.83903E-02	1.54317E 00	8.03196E-02	3.55570E-02
7.98443E-01	6.59166E-02	1.58238E 00	8.59663E-02	3.68914E-02
8.26975E-01	7.39999E-02	1.62471E 00	9.23438E-02	3.83587E-02
8.55506E-01	8.27990E-02	1.67091E 00	9.96419E-02	3.99912E-02
8.84038E-01	9.25653E-02	1.72213E 00	1.08155E-01	4.18390E-02
9.00348E-01	9.87363E-02	1.75437E 00	1.13746E-01	4.30226E-02
9.15325E-01	1.04907E-01	1.78645E 00	1.19491E-01	4.42160E-02
9.28974E-01	1.11578E-01	1.81832E 00	1.25382E-01	4.54171E-02
9.41307E-01	1.17249E-01	1.84995E 00	1.31478E-01	4.66241E-02
9.52344E-01	1.23420E-01	1.88129E 00	1.37559E-01	4.78347E-02
9.62112E-01	1.29591E-01	1.91230E 00	1.43825E-01	4.90471E-02
9.70643E-01	1.35762E-01	1.94294E 00	1.50195E-01	5.02593E-02
9.77978E-01	1.41933E-01	1.97319E 00	1.56656E-01	5.14694E-02
9.84161E-01	1.48104E-01	2.00300E 00	1.63199E-01	5.26756E-02
9.89240E-01	1.54275E-01	2.03236E 00	1.69810E-01	5.38762E-02
9.93269E-01	1.60446E-01	2.06124E 00	1.76479E-01	5.50696E-02
9.96301E-01	1.66617E-01	2.08961E 00	1.83193E-01	5.62540E-02
9.98395E-01	1.72789E-01	2.11746E 00	1.89941E-01	5.74282E-02
9.99609E-01	1.78960E-01	2.14477E 00	1.96712E-01	5.85906E-02
1.00000E 00	1.85131E-01	2.17153E 00	2.03493E-01	5.97400E-02

## HOTER 64-40

Table 6

Detonation Wave for TNT ( $\rho_0 = 1.45 \text{ gm/cc}$ )

DISTANCE X/RADIUS	VELOCITY CM/USEC	DENSITY GM/CC	PRESSURE MEGABARS	ENERGY DENS (MEG-CC)/GM
0.	0.	1.14661E 00	3.80136E-02	2.82422E-02
4.53543E-02	0.	1.14661E 00	3.80136E-02	2.82422E-02
9.07087E-02	0.	1.14661E 00	3.80136E-02	2.82422E-02
1.36063E-01	0.	1.14661E 00	3.80136E-02	2.82422E-02
1.81417E-01	0.	1.14661E 00	3.80136E-02	2.82422E-02
2.26772E-01	0.	1.14661E 00	3.80136E-02	2.82422E-02
2.72126E-01	0.	1.14661E 00	3.80136E-02	2.82422E-02
3.17480E-01	0.	1.14661E 00	3.80136E-02	2.82422E-02
3.62835E-01	0.	1.14661E 00	3.80136E-02	2.82422E-02
4.08189E-01	0.	1.14661E 00	3.80136E-02	2.82422E-02
4.53543E-01	1.71616E-04	1.14661E 00	3.80136E-02	2.82422E-02
4.82184E-01	2.46944E-03	1.15583E 00	3.88149E-02	2.85194E-02
5.10825E-01	6.05588E-03	1.17677E 00	4.61368E-02	2.89452E-02
5.39465E-01	1.02157E-02	1.18871E 00	4.17629E-02	2.94732E-02
5.58106E-01	1.47906E-02	1.20906E 00	4.36576E-02	3.00777E-02
5.96747E-01	1.97000E-02	1.23149E 00	4.58094E-02	3.07513E-02
6.25387E-01	2.49011E-02	1.25561E 00	4.82196E-02	3.14905E-02
6.54028E-01	3.03746E-02	1.28194E 00	5.08989E-02	3.22947E-02
6.82669E-01	3.61180E-02	1.30986E 00	5.38663E-02	3.31653E-02
7.11309E-01	4.21442E-02	1.33962E 00	5.71498E-02	3.41063E-02
7.39950E-01	4.84808E-02	1.37133E 00	6.07893E-02	3.51237E-02
7.68590E-01	5.51725E-02	1.40519E 00	6.48353E-02	3.62268E-02
7.97231E-01	6.22864E-02	1.44149E 00	6.93640E-02	3.74287E-02
8.25872E-01	6.99216E-02	1.48069E 00	7.44777E-02	3.87486E-02
8.54512E-01	7.82267E-02	1.52345E 00	8.03280E-02	4.02149E-02
8.83153E-01	8.74366E-02	1.57083E 00	8.71497E-02	4.18718E-02
8.99565E-01	9.32657E-02	1.60073E 00	9.16390E-02	4.29342E-02
9.14641E-01	9.90948E-02	1.63048E 00	9.62536E-02	4.40050E-02
9.28386E-01	1.04924E-01	1.66006E 00	1.00986E-01	4.50622E-02
9.40811E-01	1.10753E-01	1.68941E 00	1.05830E-01	4.61642E-02
9.51934E-01	1.16582E-01	1.71850E 00	1.10775E-01	4.72491E-02
9.61780E-01	1.22411E-01	1.74728E 00	1.15814E-01	4.83350E-02
9.70383E-01	1.28240E-01	1.77574E 00	1.20937E-01	4.94204E-02
9.77781E-01	1.34069E-01	1.80383E 00	1.26136E-01	5.05344E-02
9.84018E-01	1.39899E-01	1.83152E 00	1.31400E-01	5.15825E-02
9.89142E-01	1.45728E-01	1.85879E 00	1.36721E-01	5.26562E-02
9.93207E-01	1.51557E-01	1.88561E 00	1.42089E-01	5.37229E-02
9.96267E-01	1.57386E-01	1.91197E 00	1.47494E-01	5.47813E-02
9.98380E-01	1.63215E-01	1.93764E 00	1.52926E-01	5.58300E-02
9.99605E-01	1.69044E-01	1.96321E 00	1.58377E-01	5.68679E-02
1.00000E 00	1.74873E-01	1.98866E 00	1.63836E-01	5.78936E-02

NOLTR 64-40

Table 7

Detonation Wave for TNT ( $\rho_0 = 1.30 \text{ gm/cc}$ )

DISTANCE X/RADIUS	VELOCITY CM/USEC	DENSITY GM/CC	PRESSURE MEGABARS	ENERGY DENS (MEG-CC)/GM
0.	0.	1.02138E 00	2.99005E-02	2.97234E-02
4.53858E-02	0.	1.02138E 00	2.99005E-02	2.97234E-02
9.07715E-02	0.	1.02138E 00	2.99005E-02	2.97234E-02
1.36157E-01	0.	1.02138E 00	2.99005E-02	2.97234E-02
1.81543E-01	0.	1.02138E 00	2.99005E-02	2.97234E-02
2.26929E-01	0.	1.02138E 00	2.99005E-02	2.97234E-02
2.72315E-01	0.	1.02138E 00	2.99005E-02	2.97234E-02
3.17700E-01	0.	1.02138E 00	2.99005E-02	2.97234E-02
3.63086E-01	0.	1.02138E 00	2.99005E-02	2.97234E-02
4.08472E-01	0.	1.02138E 00	2.99005E-02	2.97234E-02
4.53858E-01	4.92819E-05	1.02138E 00	2.99005E-02	2.97234E-02
4.82401E-01	2.52551E-03	1.03098E 00	3.06106E-02	2.99994E-02
5.10944E-01	5.93218E-03	1.04474E 00	3.16466E-02	3.03970E-02
5.39487E-01	9.86073E-03	1.06118E 00	3.29137E-02	3.08755E-02
5.68030E-01	1.41695E-02	1.07977E 00	3.43860E-02	3.14212E-02
5.96573E-01	1.87854E-02	1.10022E 00	3.60553E-02	3.20276E-02
6.25116E-01	2.36693E-02	1.12239E 00	3.79229E-02	3.26914E-02
6.53659E-01	2.88034E-02	1.14619E 00	3.99970E-02	3.34120E-02
6.82202E-01	3.41857E-02	1.17161E 00	4.22921E-02	3.41905E-02
7.10745E-01	3.98276E-02	1.19870E 00	4.48297E-02	3.50302E-02
7.39288E-01	4.57546E-02	1.22754E 00	4.76396E-02	3.59362E-02
7.67831E-01	5.20076E-02	1.25833E 00	5.07624E-02	3.69165E-02
7.96374E-01	5.86481E-02	1.29133E 00	5.42537E-02	3.79822E-02
8.24917E-01	6.57666E-02	1.32694E 00	5.81922E-02	3.91499E-02
8.53460E-01	7.24985E-02	1.36576E 00	6.26922E-02	4.04437E-02
8.82003E-01	8.20569E-02	1.40873E 00	6.79310E-02	4.19014E-02
8.98544E-01	8.75274E-02	1.43613E 00	7.14140E-02	4.28445E-02
9.13748E-01	9.29978E-02	1.46341E 00	7.49958E-02	4.37944E-02
9.27617E-01	9.84683E-02	1.49053E 00	7.86728E-02	4.47496E-02
9.40161E-01	1.03939E-01	1.51746E 00	8.24330E-02	4.57084E-02
9.51395E-01	1.09409E-01	1.54416E 00	8.62759E-02	4.66693E-02
9.61344E-01	1.14880E-01	1.57059E 00	9.01929E-02	4.76306E-02
9.70040E-01	1.20350E-01	1.59672E 00	9.41760E-02	4.85909E-02
9.77520E-01	1.25821E-01	1.62251E 00	9.82206E-02	4.95486E-02
9.83828E-01	1.31291E-01	1.64795E 00	1.02317E-01	5.05024E-02
9.89012E-01	1.36762E-01	1.67301E 00	1.06457E-01	5.14508E-02
9.93125E-01	1.42232E-01	1.69765E 00	1.10635E-01	5.23926E-02
9.96222E-01	1.47702E-01	1.72187E 00	1.14842E-01	5.33266E-02
9.98361E-01	1.53173E-01	1.74565E 00	1.19072E-01	5.42515E-02
9.99600E-01	1.58643E-01	1.76896E 00	1.23316E-01	5.51663E-02
1.00000E 00	1.64114E-01	1.79180E 00	1.27567E-01	5.60700E-02

Table 8

Detonation Wave for TNT ( $\rho_0 = 1.14$  gm/cc)

DISTANCE X/RADIUS	VELOCITY CM/USEC	DENSITY GM/CC	PRESSURE MEGABARS	ENERGY DENS (MEG-CC)/GM
0.	0.	8.88453E-01	2.26994E-02	3.11573E-02
4.52631E-02	0.	8.88453E-01	2.26994E-02	3.11573E-02
9.05262E-02	0.	8.88453E-01	2.26994E-02	3.11573E-02
1.35789E-01	0.	8.88453E-01	2.26994E-02	3.11573E-02
1.81052E-01	0.	8.88453E-01	2.26994E-02	3.11573E-02
2.26316E-01	0.	8.88453E-01	2.26994E-02	3.11573E-02
2.71579E-01	0.	8.88453E-01	2.26994E-02	3.11573E-02
3.16842E-01	0.	8.88453E-01	2.26994E-02	3.11573E-02
3.62105E-01	0.	8.88453E-01	2.26994E-02	3.11573E-02
4.07368E-01	0.	8.88453E-01	2.26994E-02	3.11573E-02
4.52631E-01	8.34041E-05	8.88453E-01	2.26994E-02	3.11573E-02
4.95417E-01	3.96314E-03	9.03023E-01	2.35974E-02	3.15776E-02
5.38202E-01	9.32355E-03	9.24314E-01	2.49510E-02	3.21966E-02
5.80988E-01	1.54886E-02	9.49978E-01	2.66488E-02	3.29503E-02
6.23774E-01	2.22482E-02	9.79245E-01	2.86750E-02	3.38200E-02
6.66560E-01	2.95268E-02	1.01182E 00	3.10453E-02	3.48008E-02
7.09345E-01	3.73264E-02	1.04769E 00	3.38000E-02	3.58970E-02
7.52131E-01	4.57134E-02	1.08713E 00	3.70077E-02	3.71218E-02
7.94917E-01	5.48296E-02	1.13074E 00	4.07783E-02	3.84999E-02
8.37703E-01	6.49320E-02	1.17961E 00	4.52926E-02	4.00747E-02
8.80488E-01	7.64985E-02	1.23582E 00	5.08731E-02	4.19256E-02
8.93143E-01	8.03235E-02	1.25438E 00	5.28096E-02	4.25463E-02
9.05047E-01	8.41484E-02	1.27290E 00	5.47890E-02	4.31702E-02
9.16197E-01	8.79733E-02	1.29136E 00	5.68095E-02	4.37967E-02
9.26595E-01	9.17982E-02	1.30975E 00	5.88694E-02	4.44253E-02
9.36243E-01	9.56232E-02	1.32804E 00	6.09666E-02	4.50553E-02
9.45147E-01	9.94481E-02	1.34622E 00	6.30992E-02	4.56862E-02
9.53316E-01	1.03273E-01	1.36429E 00	6.52651E-02	4.63173E-02
9.60760E-01	1.07098E-01	1.38222E 00	6.74621E-02	4.69482E-02
9.67491E-01	1.10923E-01	1.40000E 00	6.96879E-02	4.75782E-02
9.73526E-01	1.14748E-01	1.41761E 00	7.19402E-02	4.82068E-02
9.78880E-01	1.18573E-01	1.43506E 00	7.42167E-02	4.88334E-02
9.83573E-01	1.22398E-01	1.45233E 00	7.65149E-02	4.94577E-02
9.87625E-01	1.26223E-01	1.46940E 00	7.88324E-02	5.00789E-02
9.91058E-01	1.30047E-01	1.48627E 00	8.11668E-02	5.06968E-02
9.93896E-01	1.33872E-01	1.50292E 00	8.35156E-02	5.13107E-02
9.96161E-01	1.37697E-01	1.51936E 00	8.58764E-02	5.19203E-02
9.97879E-01	1.41522E-01	1.53557E 00	8.82467E-02	5.25252E-02
9.99074E-01	1.45347E-01	1.55155E 00	9.06240E-02	5.31249E-02
9.99773E-01	1.49172E-01	1.56729E 00	9.30061E-02	5.37191E-02
1.00000E 00	1.52997E-01	1.58278E 00	9.53905E-02	5.43073E-02

## NOLTR 64-40

Table 9

Detonation Wave for TNT ( $\rho_0 = 1.00 \text{ gm/cc}$ )

DISTANCE X/RADIUS	VELOCITY CM/JSEC	DENSITY GM/CC	PRESSURE MIGABARS	ENERGY DENS (MEG-CC)/GM
0.	0.	7.72358E-01	1.74758E-02	3.22656E-02
4.46409E-02	0.	7.72358E-01	1.74758E-02	3.22656E-02
8.92817E-02	0.	7.72358E-01	1.74758E-02	3.22656E-02
1.33923E-01	0.	7.72358E-01	1.74758E-02	3.22656E-02
1.78563E-01	0.	7.72358E-01	1.74758E-02	3.22656E-02
2.23204E-01	0.	7.72358E-01	1.74758E-02	3.22656E-02
2.67845E-01	0.	7.72358E-01	1.74758E-02	3.22656E-02
3.12486E-01	0.	7.72358E-01	1.74758E-02	3.22656E-02
3.57127E-01	0.	7.72358E-01	1.74758E-02	3.22656E-02
4.01768E-01	0.	7.72358E-01	1.74758E-02	3.22656E-02
4.49272E-01	1.28491E-04	7.72358E-01	1.74758E-02	3.22656E-02
4.77912E-01	2.15273E-03	7.79481E-01	1.78420E-02	3.24745E-02
5.05551E-01	5.16212E-03	7.90505E-01	1.84196E-02	3.27988E-02
5.35190E-01	8.63988E-03	8.03724E-01	1.91259E-02	3.31893E-02
5.63830E-01	1.24547E-02	8.18707E-01	1.99486E-02	3.36341E-02
5.92469E-01	1.65385E-02	8.35218E-01	2.08816E-02	3.41269E-02
5.21109E-01	2.06542E-02	8.53127E-01	2.19252E-02	3.46647E-02
6.49748E-01	2.53841E-02	8.72308E-01	2.38388E-02	3.52463E-02
6.78387E-01	3.01246E-02	8.92925E-01	2.43650E-02	3.58722E-02
7.07027E-01	3.50840E-02	9.14832E-01	2.57844E-02	3.65443E-02
7.35666E-01	4.02828E-02	9.38170E-01	2.73462E-02	3.72663E-02
7.64305E-01	4.57547E-02	9.63075E-01	2.90844E-02	3.80437E-02
7.92945E-01	5.15505E-02	9.89701E-01	3.10251E-02	3.88846E-02
8.21584E-01	5.77454E-02	1.01854E-00	3.32137E-02	3.98009E-02
8.50223E-01	6.44517E-02	1.04984E-00	3.57028E-02	4.08101E-02
8.78863E-01	7.13447E-02	1.08452E-00	3.85960E-02	4.19395E-02
8.95739E-01	7.66343E-02	1.10694E-00	4.05481E-02	4.26784E-02
9.11281E-01	8.14240E-02	1.12931E-00	4.25581E-02	4.34216E-02
9.25483E-01	8.62136E-02	1.15157E-00	4.46229E-02	4.41678E-02
9.38347E-01	9.10322E-02	1.17371E-00	4.67391E-02	4.49156E-02
9.49885E-01	9.57929E-02	1.19567E-00	4.89031E-02	4.56640E-02
9.60118E-01	1.00583E-01	1.21744E-00	5.11110E-02	4.64116E-02
9.69072E-01	1.05372E-01	1.23898E-00	5.33587E-02	4.71573E-02
9.76782E-01	1.10162E-01	1.26020E-00	5.56419E-02	4.78999E-02
9.83290E-01	1.14951E-01	1.28126E-00	5.79564E-02	4.86384E-02
9.88643E-01	1.19741E-01	1.30195E-00	6.02977E-02	4.93716E-02
9.92892E-01	1.24531E-01	1.32231E-00	6.26612E-02	5.00986E-02
9.96093E-01	1.29320E-01	1.34233E-00	6.50424E-02	5.08185E-02
9.98304E-01	1.34110E-01	1.36198E-00	6.74369E-02	5.15303E-02
9.99586E-01	1.39900E-01	1.38125E-00	6.98401E-02	5.22333E-02
1.00000E-00	1.43689E-01	1.40012E-00	7.22476E-02	5.29266E-02

ARMY

COPIES

CHIEF OF RESEARCH AND DEVELOPMENT, D/A  
ATTN. ATOMIC DIVISION  
WASHINGTON, D.C. 20310

1

CHIEF OF ENGINEERS, D/A  
ATTN. ENGCW-NE, ENGTE-E, ENGMC-E  
WASHINGTON, D.C. 20310

3

COMMANDING GENERAL  
U.S. ARMY MATERIEL COMMAND  
ATTN. AMCRU-DE-N  
WASHINGTON, D.C. 20310

2

COMMANDING GENERAL  
U.S. CONTINENTAL ARMY COMMAND  
FT MONROE, VA. 23351

1

PRESIDENT  
U.S. ARMY AIR DEFENSE BOARD  
FT BLISS, TEXAS 79906

1

COMMANDANT  
COMMAND AND GENERAL STAFF COLLEGE  
ATTN. ARCHIVES  
FT LEAVENWORTH, KANSAS 66027

1

COMMANDANT  
U.S. ARMY AIR DEFENSE SCHOOL  
ATTN. COMMAND AND STAFF DEPT  
FT BLISS, TEXAS 79906

1

DIRECTOR  
SPECIAL WEAPONS DEVELOPMENT  
HQ CDC  
ATTN. CHESTER I. PETERSON  
FT BLISS, TEXAS 79906

1

COMMANDING GENERAL  
ABERDEEN PROVING GROUND  
ATTN. BRL FOR DIRECTOR, J. J. MESZAROS  
W. J. TAYLOR, R. E. SHEAR  
ABERDEEN, MARYLAND 21005

2

2

COMMANDING GENERAL  
THE ENGINEER CENTER  
ATTN. ASST COMMANDANT, ENGINEER SCHOOL  
FT BELVOIR, VIRGINIA 22060

1

DIRECTOR  
U.S. ARMY RESEARCH AND DEVELOPMENT LABORATORY  
ATTN. CHIEF, TECH SUPPORT BRANCH  
FT BELVOIR, VIRGINIA 22060

1

COMMANDING OFFICER  
PICATINNY ARSENAL  
ATTN. SMUPA-G,-W,-VL,-VE,-VC,-DD,-DR,-DR-4,-DW,-TX,-TW,-V 12  
DOVER, N.J. 07801

DIRECTOR  
U.S. ARMY CORPS OF ENGINEERS  
WATERWAYS EXPERIMENT STATION  
ATTN. LIBRARY, JOHN STRANGE  
VICKSBURG, MISSISSIPPI 39180

2

COMMANDING GENERAL  
USA MISSILE COMMAND  
HUNTSVILLE, ALABAMA 35801

1

NAVY

CHIEF OF NAVAL OPERATIONS, ND  
ATTN. OP-75  
ATTN. OP-03EG  
WASHINGTON, D.C. 20350

2  
1

DIRECTOR OF NAVAL INTELLIGENCE, ND  
ATTN. OP-922V  
WASHINGTON, D.C. 20350

1

SPECIAL PROJECTS OFFICE, ND  
ATTN. SP-272  
WASHINGTON, D.C. 20360

1

CHIEF  
BUREAU OF NAVAL WEAPONS, ND  
ATTN. DLI-3  
ATTN. RUME-11, RUME-4, RRRE-5  
WASHINGTON, D.C. 20360

4  
3

CHIEF  
BUREAU OF SHIPS, ND  
ATTN. CODE 372, CODE 423  
WASHINGTON, D.C. 20360

2

CHIEF  
BUREAU OF YARDS AND DOCKS, ND  
ATTN. CODE D-400, CODE D-440  
WASHINGTON, D.C. 20370

2

CHIEF OF NAVAL RESEARCH, ND  
ATTN. CODE 811  
ATTN. CODE 493, CODE 418, CODE 104  
WASHINGTON, D.C. 20390

2  
3

COMMANDER-IN-CHIEF  
U.S. PACIFIC FLEET, FPO  
SAN FRANCISCO, CALIF. 96601

1

COMMANDER-IN-CHIEF  
U.S. ATLANTIC FLEET  
U.S. NAVAL BASE  
NORFOLK, VIRGINIA 23500

1

COMMANDANT OF THE MARINE CORPS, ND  
ATTN. CODE A03H  
WASHINGTON, D.C. 20350

4

PRESIDENT  
U.S. NAVAL WAR COLLEGE  
NEWPORT, R.I. 02840

1

SUPERINTENDENT  
U.S. NAVAL POSTGRADUATE SCHOOL  
MONTEREY, CALIFORNIA 93940

1

COMMANDING OFFICER  
NUCLEAR WEAPONS TRAINING CENTER, ATLANTIC  
ATTN. NUCLEAR WARFARE DEPARTMENT  
NAVAL BASE  
NORFOLK, VIRGINIA 23500

1

COMMANDING OFFICER  
U.S. NAVAL SCHOOLS COMMAND  
U.S. NAVAL STATION  
TREASURE ISLAND  
SAN FRANCISCO, CALIFORNIA 94100

1

COMMANDING OFFICER  
NUCLEAR WEAPONS TRAINING CENTER, PACIFIC  
NAVAL STATION  
NORTH ISLAND  
SAN DIEGO, CALIFORNIA 92100

2

COMMANDING OFFICER  
U.S. NAVAL DAMAGE CONTROL TRAINING CENTER  
ATTN. ABC DEFENSE COURSE  
NAVAL BASE  
PHILADELPHIA, PA. 19100

1

COMMANDER  
U.S. NAVAL ORDNANCE TEST STATION  
ATTN. LIBRARY, R. E. BOYER, DR. MALLORY  
CHINA LAKE, CALIFORNIA 96105

3

COMMANDING OFFICER AND DIRECTOR  
U.S. NAVAL CIVIL ENGINEERING LABORATORY  
ATTN. CODE L31  
PORT HUENEME, CALIFORNIA 93041

1

DIRECTOR  
U.S. NAVAL RESEARCH LABORATORY  
ATTN. LIBRARY  
DR. LOUIS F. DRUMMETER  
WASHINGTON, D.C. 20390

1

COMMANDING OFFICER AND DIRECTOR  
NAVY ELECTRONICS LABORATORY  
SAN DIEGO, CALIFORNIA 92100

1

COMMANDING OFFICER  
U.S. NAVAL RADIOLOGICAL DEFENSE LABORATORY  
ATTN. TECH INFO DIV  
SAN FRANCISCO, CALIFORNIA 94129

1

COMMANDING OFFICER AND DIRECTOR  
DAVID W. TAYLOR MODEL BASIN  
ATTN. LIBRARY  
WASHINGTON, D.C. 20007

1

UNDERWATER EXPLOSIONS RESEARCH DIVISION  
DAVID W. TAYLOR MODEL BASIN  
NORFOLK NAVAL SHIPYARD  
PORTSMOUTH, VIRGINIA 23700

1

COMMANDER  
U.S. NAVAL WEAPONS EVALUATION FACILITY  
ATTN. MR. EARL B. MASSINGILL, JR.  
KIRTLAND AFB  
NEW MEXICO

1

COMMANDING OFFICER  
U.S. NAVAL PROPELLANT PLANT  
ATTN. TECHNICAL LIBRARY  
INDIAN HEAD, MARYLAND

1

COMMANDER  
U. S. NAVAL WEAPONS LABORATORY  
ATTN. TERMINAL BALLISTICS DEPARTMENT  
TECHNICAL LIBRARY  
DAHLGREN, VIRGINIA

2

COMMANDING OFFICER  
U.S. NAVAL ORDNANCE LABORATORY  
CORONA, CALIFORNIA

1

AIR FORCE

HQ, USAF  
(AFRNE)  
WASHINGTON, D.C. 20330

1

DEPUTY CHIEF OF STAFF  
PLANS AND PROGRAMS, HQ USAF  
ATTN. WAR PLANS DIVISION  
WASHINGTON, D.C. 20330

1

DIRECTOR OF RESEARCH AND DEVELOPMENT  
DCS/D, HQ USAF  
ATTN. GUIDANCE AND WEAPONS DIVISION  
WASHINGTON, D.C. 20330

1

AIR FORCE INTELLIGENCE CENTER  
HQ USAF, ACS/I (AFCIN-3K2)  
WASHINGTON, D.C. 22212

1

COMMANDER-IN-CHIEF  
STRATEGIC AIR COMMAND  
ATTN. OAWS  
OFFUTT AFB, NEBRASKA 68113

1

COMMANDER  
TACTICAL AIR COMMAND  
ATTN. DOCUMENT SECURITY BRANCH  
LANGLEY AFB, VIRGINIA 23365

1

ASD  
WRIGHT-PATTERSON AFB  
OHIO 45433

1

COMMANDER  
AIR FORCE LOGISTICS COMMAND  
WRIGHT-PATTERSON AFB, OHIO 45433

2

AFSC, ANDREWS AIR FORCE BASE  
ATTN. RDRWA  
WASHINGTON, D.C. 20331

1

DIRECTOR  
AIR UNIVERSITY LIBRARY  
MAXWELL AFB, ALABAMA 36112

2

AFCRL, L. G. HANSCOM FIELD  
ATTN. CRQST, H. P. GAUVIN  
DR. N. ROSENBERG  
BEDFORD, MASSACHUSETTS 01731

2  
1

AFWL  
ATTN. CAPT. C.M. GILLESPIE  
KIRTLAND AFB, NEW MEXICO 87117

4

COMMANDANT  
INSTITUTE OF TECHNOLOGY  
ATTN. MCLI-ITRIDL  
WRIGHT-PATTERSON AFB, OHIO 45433

1

BSD  
NORTON AFB  
CALIFORNIA 92409

1

DIRECTOR  
USAF PROJECT RAND  
VIA. U.S. AIR FORCE LIAISON OFFICE  
THE RAND CORPORATION  
ATTN. DR. H. L. BRODE  
1700 MAIN STREET  
SANTA MONICA, CALIFORNIA 90400

1

DIRECTOR OF CIVIL ENGINEERING  
HQ USAF  
ATTN. AFOCE  
WASHINGTON, D.C. 20330

1

COMMANDING GENERAL  
WHITE SANDS MISSILE RANGE  
WHITE SANDS, NEW MEXICO

1

OTHERS

DIRECTOR OF DEFENSE RESEARCH AND ENGINEERING  
ATTN. TECH LIBRARY  
WASHINGTON, D.C. 20330

1

U.S. DOCUMENTS OFFICER  
OFFICE OF THE UNITED STATES NATIONAL  
MILITARY REPRESENTATIVE, SHAPE  
APO 55  
NEW YORK, N.Y. 10000

1

COMMANDER-IN-CHIEF, PACIFIC  
FLEET POST OFFICE  
SAN FRANCISCO, CALIFORNIA 94129

1

DIRECTOR  
WEAPONS SYSTEMS EVALUATION GROUP, OSD  
ROOM 1E880  
THE PENTAGON  
WASHINGTON, D.C. 20301

1

COMMANDANT  
ARMED FORCES STAFF COLLEGE  
ATTN. LIBRARY  
NORFOLK, VIRGINIA 23500

1

CHIEF  
WEAPONS TEST DIVISION  
DEFENSE ATOMIC SUPPORT AGENCY  
ATTN. FCWT, FCTG,  
SANDIA BASE  
ALBUQUERQUE, NEW MEXICO

1

2

DIRECTOR  
DEFENSE ATOMIC SUPPORT AGENCY  
WASHINGTON, D.C. 20301

5

COMMANDANT  
ARMY WAR COLLEGE  
ATTN. LIBRARY  
CARLISLE BARRACKS  
PENNSYLVANIA 17013

1

COMMANDANT  
NATIONAL WAR COLLEGE  
ATTN. CLASS REC. LIBRARY  
WASHINGTON, D.C. 20390

1

COMMANDANT  
THE INDUSTRIAL COLLEGE OF THE ARMED FORCES  
FT MCNAIR  
WASHINGTON, D.C. 20310

1

OFFICER-IN-CHARGE  
U.S. NAVAL SCHOOL  
CIVIL ENGINEERING CORPS OFFICERS  
NAVAL CONSTRUCTION BATTALION  
PORT HUENEME, CALIFORNIA 93041

1

LOS ALAMOS SCIENTIFIC LABORATORY  
ATTN. DR. A. C. GRAVES  
DR. W. C. DAVIES  
P.O. BOX 1663  
LOS ALAMOS, NEW MEXICO 87544

1

SCIENTIFIC AND TECHNICAL INFORMATION FACILITY  
ATTN. NASA REPRESENTATIVE  
P. O. BOX 5700  
BETHESDA, MARYLAND 20000

1

LANGLEY RESEARCH CENTER  
NASA, LANGLEY FIELD  
ATTN. MR. PHILIP DONELY  
HAMPTON, VIRGINIA 23365

1

CHIEF  
CLASSIFIED TECHNICAL LIBRARY  
TECHNICAL INFORMATION SERVICE  
U.S. ATOMIC ENERGY COMMISSION  
ATTN. MRS. JEAN O LEARY  
WASHINGTON, D.C. 20545

1

MANAGER  
ALBUQUERQUE OPERATIONS OFFICE  
U.S. ATOMIC ENERGY COMMISSION  
P. O. BOX 5400  
ALBUQUERQUE, NEW MEXICO 87100

1

PRESIDENT  
SANDIA CORPORATION, SANDIA BASE  
ATTN. DR. M. L. MERRITT, W. B. BENEDICK  
W. ROBERTS, J. W. REED, DR. C. BROYLES  
ALBUQUERQUE, NEW MEXICO 87115

2

3

UNITED RESEARCH SERVICES  
ATTN. MR. KENNETH KAPLAN  
1811 TROUSDALE DRIVE  
BURLINGAME, CALIFORNIA 94010

1

DEPARTMENT OF PHYSICS  
STANFORD RESEARCH INSTITUTE  
ATTN. MR. FRED M. SAUER, DR. GEO. DUVALL  
MENLO PARK, CALIFORNIA

2

SPACE TECHNOLOGY LABORATORIES, INC.  
ATTN. DR. LEON  
5500 WEST EL SEGUNDO BLVD.  
LOS ANGELES, CALIFORNIA 90000

1

PRESIDENT  
KAMAN NUCLEAR  
ATTN. DR. FRANK SHELTON  
COLORADO SPRINGS, COLORADO 80900

1

E. O. LAWRENCE RADIATION LABORATORY  
UNIVERSITY OF CALIFORNIA  
ATTN. TECH INFO DIV  
P. O. BOX 808  
LIVERMORE, CALIFORNIA 94550

2

NATIONAL AERONAUTICS AND SPACE ADMINISTRATION  
MAN-SPACECRAFT CENTER  
SPACE TECHNOLOGY DIVISION  
ATTN. MR. R. F. FLETCHER  
BOX 1537  
HOUSTON, TEXAS 77000

1

IIT RESEARCH INSTITUTE  
ILLINOIS INSTITUTE OF TECHNOLOGY  
10 WEST 35TH STREET  
CHICAGO 16, ILLINOIS

1

INSTITUTE FOR THE STUDY OF RATE PROCESSES  
UNIVERSITY OF UTAH  
ATTN. DR. M. A. COOK  
SALT LAKE CITY, UTAH

1

DENVER RESEARCH INSTITUTE  
MECHANICS DIVISION  
UNIVERSITY OF DENVER  
ATTN. MR. RODNEY F. RECHT  
(CONTRACT N60921-7013)  
DENVER 10, COLORADO  
VIA. ONRRR  
LAWRENCE, KANSAS

2

DIRECTOR  
NEW MEXICO INSTITUTE OF MINING TECHNOLOGY  
ATTN. DR. M. KEMPTON  
SOCORRO, NEW MEXICO  
VIA. BUWEPS (RUME-4)  
WASHINGTON, D. C. 20360

1

DIRECTOR  
APPLIED PHYSICS LABORATORY  
JOHNS HOPKINS UNIVERSITY  
ATTN. TECH LIBRARY  
8621 GEORGIA AVENUE  
SILVER SPRING, MARYLAND  
VIA. BUWEPS (RMMO-5)  
WASHINGTON, D. C. 20360

1

DIRECTOR  
U.S. BUREAU OF MINES  
DIVISION OF EXPLOSIVE TECHNOLOGY  
ATTN. DR. ROBERT W. VAN DOLAH  
4800 FORBES STREET  
PITTSBURGH 13, PENNSYLVANIA

1

CHAIRMAN  
ARMED SERVICES EXPLOSIVES SAFETY BOARD  
ATTN. MR. R. G. PERKINS  
BLDG T-7  
GRAVELLY POINT  
WASHINGTON, D. C.

1

SCIENCE COMMUNICATION, INC.  
ATTN. MR. D. O. MYATT  
1079 WISCONSIN AVENUE, N. W.  
WASHINGTON, D.C. 20007

1

EDGERTON, GERMESHAUSEN, AND GRIER, INC.  
ATTN. D. F. HANSEN  
160 BROOKLINE AVENUE  
BOSTON 15, MASSACHUSETTS 02129

1

OFFICE OF TECHNICAL SERVICES  
DEPARTMENT OF COMMERCE  
WASHINGTON 25, D. C.

100

PHYSICS INTERNATIONAL CO.  
ATTN. DR. W. BIRNBAUM  
2229 4TH STREET  
BERKELEY, CALIFORNIA 94700

1

ASTROPHYSICS RESEARCH CORPORATION  
ATTN. DR. A. REIFMAN  
10889 WILSHIRE BLVD.  
LOS ANGELES, CALIFORNIA 94700

1

PHILCO CORPORATION, AERONUTRONIC DIVISION AND PHILCO RESEARCH LABS.  
ATTN. DR. R. A. GRANDEY  
FORD ROAD  
NEWPORT BEACH, CALIFORNIA 92660

1

HOUSTON RESEARCH INSTITUTE, INCORPORATED  
ATTN. CLARK GOODMAN  
5417 CRAWFORD STREET  
HOUSTON, TEXAS 77004

1

DDC  
CAMERON STATION  
ATTN. TISIA-21  
ALEXANDRIA, VIRGINIA

20

FORRESTAL RESEARCH CENTER LIBRARY  
AERONAUTICAL SCIENCES BLDG.  
PRINCETON UNIVERSITY  
ATTN. LIBRARIAN  
(FOR DR. WALKER BLEAKNEY)  
PRINCETON, N.J. 08540

1

UNCLASSIFIED

Security Classification

DOCUMENT CONTROL DATA - R&D

(Security classification of title, body of abstract and in-text & any column to be entered when the were "SECRET" or "TOP SECRET")

1 ORIGINATING ACTIVITY (Corporate author) <b>U. S. NAVAL ORDNANCE LABORATORY WHITE OAK, SILVER SPRING, MARYLAND</b>	2A REPORT SEC R&D CLASSIFICATION <b>UNCLASSIFIED</b>	
	2B GROUP <b>N/A</b>	
3 REPORT TITLE <b>THE FLOW FIELD BEHIND A SPHERICAL DETONATION IN TNT USING THE LANDAU-STANYUKOVICH EQUATION OF STATE FOR DETONATION PRODUCTS</b>		
4 DESCRIPTIVE NOTES (Type of report and inclusive dates) <b>Final</b>		
5 AUTHOR(S) (Last name, first name, initial) <b>Lutzky, Morton</b>		
6 REPORT DATE <b>10 December 1964</b>	7A TOTAL NO OF PAGES <b>27</b>	7B NO OF REFS <b>13</b>
8 CONTRACT OR GRANT NO	9A ORIGINATOR'S REPORT NUMBER(S) <b>NOLTR 64-40</b>	
6 PROJECT NO <b>Task NOL-428</b>	9B OTHER REPORT NO'S (An - other numbers that may be assigned to this report)	
c		
d		
10 AVAILABILITY LIMITATION NOTICES Released to DDC without restriction.		
11 SUPPLEMENTARY NOTES	12 SPONSORING MILITARY ACTIVITIES <b>Defense Atomic Support Agency The Pentagon, Washington, D.C.</b>	
13 ABSTRACT <p>Calculations have been made of the flow field in the isentropic region behind a detonation wave in TNT, using the Landau-Stanyukovich equation of state for the detonation products (as described by Zeldovich and Kompaneets). Adjustable constants in this equation have been evaluated by imposing ideal gas behavior on the detonation products in the large expansion (low density) limit, and by fitting to an experimental curve of detonation velocity versus loading density. Calculated values of Chapman-Jouguet variables correspond fairly well with experimental values at various loading densities, with the exception of the temperatures, which seem to be far too low. This is connected with the fact that the theory predicts an upper limit to the loading density at which an explosive will detonate; at this point the thermal energy vanishes and only the elastic energy contributes to the energy of detonation.</p>		

DD FORM 1 JAN 64 1473

Security Classification

## Security Classification

14 KEY WORDS	LINK A		LINK B		LINK C	
	ROLE	WT	ROLE	WT	ROLE	WT
Detonation Products High Explosives Equation of State Taylor Wave TNT						
INSTRUCTIONS						
1. ORIGINATING ACTIVITY Enter the name and address of the contractor subcontractor, grantee, Department of Defense activity or other organization (corporate author) issuing the report.	imposed by security classification, using standard statements such as					
2a. REPORT SECURITY CLASSIFICATION Enter the overall security classification of the report. Indicate whether "Restricted Data" is included. Marking is to be in accordance with appropriate security regulations.	<ul style="list-style-type: none"> <li>(1) "Qualified requesters may obtain copies of this report from DDC."</li> <li>(2) "Foreign announcement and dissemination of this report by DDC is not authorized."</li> <li>(3) "U S Government agencies may obtain copies of this report directly from DDC. Other qualified DDC users shall request through _____."</li> </ul>					
2b. GROUP Automatic downgrading is specified in DoD Directive 5200.10 and Armed Forces Industrial Manual. Enter the group number. Also, when applicable, show that optional markings have been used for Group 3 and Group 4 as authorized.	<ul style="list-style-type: none"> <li>(4) "U S military agencies may obtain copies of this report directly from DDC. Other qualified users shall request through _____."</li> <li>(5) "All distribution of this report is controlled. Qualified DDC users shall request through _____."</li> </ul>					
3. REPORT TITLE Enter the complete report title in all capital letters. Titles in all cases should be unclassified. If a meaningful title cannot be selected without classification, show title classification in all capitals in parentheses immediately following the title.	If the report has been furnished to the Office of Technical Services, Department of Commerce, for sale to the public, indicate this fact and enter the price, if known.					
4. DESCRIPTIVE NOTES If appropriate, enter the type of report, e.g., interim, progress, summary, annual, or final. Give the inclusive dates when a specific reporting period is covered.	11. SUPPLEMENTARY NOTES Use for additional explanatory notes.					
5. AUTHOR(S) Enter the name(s) of author(s) as shown on or in the report. Enter last name, first name, middle initial. If military, show rank and branch of service. The name of the principal author is an absolute minimum requirement.	12. SPONSORING MILITARY ACTIVITY Enter the name of the departmental project office or laboratory sponsoring (paying for) the research and development. Include address.					
6. REPORT DATE Enter the date of the report as day, month, year, or month, year. If more than one date appears on the report, use date of publication.	13. ABSTRACT Enter an abstract giving a brief and factual summary of the document indicative of the report, even though it may also appear elsewhere in the body of the technical report. If additional space is required, a continuation sheet shall be attached.					
7a. TOTAL NUMBER OF PAGES The total page count should follow normal pagination procedures, i.e., enter the number of pages containing information.	<p style="text-align: center;">It is highly desirable that the abstract of classified reports be unclassified. Each paragraph of the abstract shall end with an indication of the military security classification of the information in the paragraph, represented as (TS) (S) (C) or (U)</p>					
7b. NUMBER OF REFERENCES Enter the total number of references cited in the report.	<p style="text-align: center;">There is no limitation on the length of the abstract. However, the suggested length is from 150 to 225 words.</p>					
8a. CONTRACT OR GRANT NUMBER If appropriate, enter the applicable number of the contract or grant under which the report was written.	14. KEY WORDS Key words are technically meaningful terms or short phrases that characterize a report and may be used as index entries for cataloging the report. Key words must be selected so that no security classification is required. Identifiers, such as equipment model designation, trade name, military project code name, geographic location, may be used as key words but will be followed by an indication of technical context. The assignment of links, roles, and weights is optional.					
8b. & 8d. PROJECT NUMBER Enter the appropriate military department identification, such as project number sub-project numbers, task number, etc.						
9a. ORIGINATOR'S REPORT NUMBER(S) Enter the official report number by which the document will be identified and controlled by the originating activity. This number must be unique to this report.						
9b. OTHER REPORT NUMBERS If the report has been assigned any other report numbers, thereby the originator or the sponsor, also enter the numbers.						
10. AVAILABILITY LIMITATION NOTES Enter any limitation on further dissemination of the report other than those						

<p>Naval Ordnance Laboratory, White Oak, Md. (NOL technical report 54-40) THE FLOW FIELD BEHIND A SPHERICAL DETONATION IN TNT USING THE LANDAU-STANYUKOVICH EQUATION OF STATE FOR DETONATION PRODUCTS, by M. Lutzky. 10 Dec. 1964. 15p. tables. NOL task 248.</p> <p>UNCLASSIFIED</p> <p>III. Project</p>	<p>Calculations have been made of the flow field in the isentropic region behind a detonation wave in TNT. Adjustable constants in this equation have been evaluated by imposing ideal gas behavior on the detonation products in the large expansion limit, and by fitting to an experimental curve of detonation velocity versus loading density. Abstract card is unclassified.</p>	<p>1. Explosives - Detonation 2. Explosions - Shock waves I. Title II. Lutzky, Morton III. Project</p> <p>Naval Ordnance Laboratory, White Oak, Md. (NOL technical report 64-40) THE FLOW FIELD BEHIND A SPHERICAL DETONATION IN TNT USING THE LANDAU-STANYUKOVICH EQUATION OF STATE FOR DETONATION PRODUCTS, by M. Lutzky. 10 Dec. 1964. 15p. tables. NOL task 248.</p> <p>UNCLASSIFIED</p>	<p>Calculations have been made of the flow field in the isentropic region behind a detonation wave in TNT. Adjustable constants in this equation have been evaluated by imposing ideal gas behavior on the detonation products in the large expansion limit, and by fitting to an experimental curve of detonation velocity versus loading density. Abstract card is unclassified.</p>
<p>1. Explosives - Detonation 2. Explosions - Shock waves I. Title II. Lutzky, Morton III. Project</p> <p>Naval Ordnance Laboratory, White Oak, Md. (NOL technical report 64-40) THE FLOW FIELD BEHIND A SPHERICAL DETONATION IN TNT USING THE LANDAU-STANYUKOVICH EQUATION OF STATE FOR DETONATION PRODUCTS, by M. Lutzky. 10 Dec. 1964. 15p. tables. NOL task 248.</p> <p>UNCLASSIFIED</p>	<p>Calculations have been made of the flow field in the isentropic region behind a detonation wave in TNT. Adjustable constants in this equation have been evaluated by imposing ideal gas behavior on the detonation products in the large expansion limit, and by fitting to an experimental curve of detonation velocity versus loading density. Abstract card is unclassified.</p>	<p>1. Explosives - Detonation 2. Explosions - Shock waves I. Title II. Lutzky, Morton III. Project</p> <p>Naval Ordnance Laboratory, White Oak, Md. (NOL technical report 64-40) THE FLOW FIELD BEHIND A SPHERICAL DETONATION IN TNT USING THE LANDAU-STANYUKOVICH EQUATION OF STATE FOR DETONATION PRODUCTS, by M. Lutzky. 10 Dec. 1964. 15p. tables. NOL task 248.</p> <p>UNCLASSIFIED</p>	<p>Calculations have been made of the flow field in the isentropic region behind a detonation wave in TNT. Adjustable constants in this equation have been evaluated by imposing ideal gas behavior on the detonation products in the large expansion limit, and by fitting to an experimental curve of detonation velocity versus loading density. Abstract card is unclassified.</p>
<p>1. Explosives - Detonation 2. Explosions - Shock waves I. Title II. Lutzky, Morton III. Project</p> <p>Naval Ordnance Laboratory, White Oak, Md. (NOL technical report 64-40) THE FLOW FIELD BEHIND A SPHERICAL DETONATION IN TNT USING THE LANDAU-STANYUKOVICH EQUATION OF STATE FOR DETONATION PRODUCTS, by M. Lutzky. 10 Dec. 1964. 15p. tables. NOL task 248.</p> <p>UNCLASSIFIED</p>	<p>Calculations have been made of the flow field in the isentropic region behind a detonation wave in TNT. Adjustable constants in this equation have been evaluated by imposing ideal gas behavior on the detonation products in the large expansion limit, and by fitting to an experimental curve of detonation velocity versus loading density. Abstract card is unclassified.</p>	<p>1. Explosives - Detonation 2. Explosions - Shock waves I. Title II. Lutzky, Morton III. Project</p> <p>Naval Ordnance Laboratory, White Oak, Md. (NOL technical report 64-40) THE FLOW FIELD BEHIND A SPHERICAL DETONATION IN TNT USING THE LANDAU-STANYUKOVICH EQUATION OF STATE FOR DETONATION PRODUCTS, by M. Lutzky. 10 Dec. 1964. 15p. tables. NOL task 248.</p> <p>UNCLASSIFIED</p>	<p>Calculations have been made of the flow field in the isentropic region behind a detonation wave in TNT. Adjustable constants in this equation have been evaluated by imposing ideal gas behavior on the detonation products in the large expansion limit, and by fitting to an experimental curve of detonation velocity versus loading density. Abstract card is unclassified.</p>